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DATA REPORT FOR A TEST PROGRAM TO STUDY TRANSONIC FLOW FIELDS ABOUT AIRCRAFT WITH APPLICATION TO EXTERNAL STORES

VOLUME V. - FLOW-FIELD SURVEY DATA FOR THE 6-PERCENT THICK WING-BODY COMBINATION

By Stanley C. Perkins, Jr., Stephen S. Stahara and Michael J. Hemsch

NEAR TR 138

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20. ABSTRACT (Continue on reverse side II necessary and identify by block number A test program was conducted to obtain measurement static pressures in the vicinity of wing-body-stor fighter-type aircraft) as well as surface pressure model. Flow velocities and static pressures were	ts of flow velocities and re model (representative of a es, forces, and moments on the

walls to provide outer flow field information. This report presents the data obtained during the test program conducted in the 4T and 16T Wind Tunnels at Arnold Engineering Development Center. The Flow-field data were obtained at Mach numbers 0.925, 0.975, and 1.025 and constitute the major part of the data. (cont

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Volume I is a summary report which gives detailed information on the test program and presents uncertainties associated with the various types of data taken in the 4T Wind Tunnel. The volume also presents tunnel-empty and Mach-number surveys, as well as tabulated force and moment and pressure data for the Mach number range 0.80 to 1.15 and angles of attack -2°, -5°, 0°, 2°, and 5°. Volumes II, III, and IV present the tabulated flowfield data for the 4-percent thick wing model at Mach numbers 0.925, 0.975 and 1.025, respectively. Volume V presents the tabulated flow-field data for the 6-percent thick wing model, and Volume VI presents data obtained for the 4-percent thick wing model in the 16T Wind Tunnel.

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DATA	1

NOMENCIATURE NEW YORK

This section provides a list of symbols which identify various aerodynamic parameters, axis designations, subscripts, and tabulated data nomenclature.

Symbols

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AAL

AAL	local upwash angle, deg; tan (WL/VM)	
Tenfers	sor brow sich is to entry school VM	· BEAG
M	Mach number sent renaus baly	
de maggarenent	radius of the body, in. down front	.04
Re/ft	free-stream Reynolds number per foot, ft-1	T.
SWL ab pulbus	local sidewash angle, deg; tan (VL/VM)	TRAT.
UL,VL,WL	local velocity components, positive along positive X, Y, and Z directions, respetft/sec	the ctively,
spoing dat MV	free-stream velocity, ft/sec	POTON
X,Y,Z		
	tunnel-fixed Cartesian coordinates with or coincident with the aircraft model nose a angle of attack, see figure 4(b)	
a , redewn a	angle of attack of the model, deg; angle body axis and tunnel axis, as defined in	
e entidade year	azimuthal angle in the Y-Z plane, deg; m from the positive Y axis as shown in fi	
	Subscripts	
alxa-yho	rd of benneraler elgos dezweble Lapot	TMB :
· 一	(ANALY) 1 ms (pso	

ponediaates, calculated from pr

Flow Field Survey Data Tabulation

local upwash angle referenced to body-axis coordinates, calculated from probe measurements, deg; tan (WL/UL)

NOMENCLATURE (Continued)

AATL Yllingb	upwash angle referenced to tunnel-axis coordinates, calculated from probe measurements, deg;
, esqlapade	tan ⁻¹ (WT/UT)
ALFA	aircraft-model angle of attack, positive nose up as seen by the pilot (nose down in tunnel), deg
CPL	local pressure coefficient calculated from probe measurements, (PL -P)/Q
DATE	calendar time at which data were recorded
, M	wind tunnel free-stream Mach number
ML	local Mach number calculated from probe measurements
P 1"31	free-stream static pressure, psfa
PART	sequential indexing number for referencing data; a constant throughout each survey
PL viscosquex	local static pressure calculated from probe measure- ments, psfa
POINT	sequential indexing number for referencing data obtained during one part; indexes each time a new set of data inputs is obtained
PT ds mao	wind tunnel free-stream total pressure, psfa
PTL lelio di	local total pressure measured by probe, psfa
Q	wind tunnel free-stream dynamic pressure, psf
REX10-6	wind tunnel free-stream unit Reynolds number, millions per foot
RUN	identifier for specific user test type
SURVEY	identifier for specific user grid-survey combination
SWL	local sidewash angle referenced to body-axis coordinates, calculated from probe measurements, deg; tan (VL/UL)
SWTL	sidewash angle referenced to tunnel-axis coordinates, calculated from probe measurements, deg; tan (VT/UT)
123日他制造工作的	

NOMENCLATURE (Concluded)

,VL,WL ,VT,WT	wind tunnel free-stream total temperature, ^o F velocity components in the body-axis X, Y, and Z directions, respectively, calculated from probe measurements, ft/sec velocity components in the tunnel-axis X, Y, and Z directions, respectively, calculated from probe measurements, ft/sec wind tunnel free-stream velocity, ft/sec local velocity calculated from probe measurements, ft/sec
,VT,WT	<pre>Z directions, respectively, calculated from probe measurements, ft/sec velocity components in the tunnel-axis X, Y, and Z directions, respectively, calculated from probe measurements, ft/sec wind tunnel free-stream velocity, ft/sec local velocity calculated from probe measurements,</pre>
	Z directions, respectively, calculated from probe measurements, ft/sec wind tunnel free-stream velocity, ft/sec local velocity calculated from probe measurements,
	local velocity calculated from probe measurements,
4	
NG	wing designation used for a specific part number
	location of the probe in the body-axis X direction
	location of the probe in the tunnel-axis X direction
	location of the probe in the body-axis Y direction
	location of the probe in the tunnel-axis Y direction
	location of the probe in the body-axis Z direction
	location of the probe in the tunnel-axis Z direction

DATA REPORT FOR A TEST PROGRAM TO STUDY TRANSONIC FLOW FIELDS ABOUT AIRCRAFT WITH APPLICATION TO EXTERNAL STORES

VOLUME V. - FLOW-FIELD SURVEY DATA FOR
THE 6-PERCENT THICK WING-BODY COMBINATION

of the bebulated date for each flow-field run. Columns cour.

1. INTRODUCTION of lack? but isidial out essential aim bas joyit

This volume of the data report presents the flow-field survey data at $M_{\infty}=0.925$, 0.975 and 1.025 for the 6-percent thick wing-body model. The data were obtained in the 4T Wind Tunnel at Arnold Engineering and Development Center. These tests, performed at a nominal Reynolds number per foot of 3.0×10^6 , are outlined in Tables I through XIV of this volume. The tabulated data are at the end of this volume beginning on page number 1.

2. DESCRIPTION OF TESTS

The details of the test hardware, coordinate systems, and data uncertainties associated with these tests, as well as an overview of the purpose and scope of the test program, are provided in Volume I of this data report. Figure 1 of this volume shows the entire grid layout used for the inner flow-field measurements. Some layouts use only a portion of this pattern, depending on Mach number and angle of attack. The general grid layout used for the outer flow-field measurements is shown in figure 2. Two different grid layouts were used depending on the angle of attack. The grid layout is designed to give outer flow-field data at a constant radial distance from the tunnel centerline for various values of θ , the azimuthal angle. A sketch of the wing-body combination is shown in figure 3.

of the axial coordinate, the

The flow field survey tests for the 6-percent thick wingbody model are summarized in Table I. Column one indicates the Mach number at which the data were obtained. Columns two through four indicate the angle of attack, the table number, and the page numbers, respectively, for each set of data.

Tables II through XIV contain the test condition grids for

each Mach number/angle of attack combination. Columns one and two indicate the page number and part number, respectively, of the tabulated data for each flow-field run. Columns four, five, and six indicate the initial and final positions and the incremental change in the axial coordinate, the body-axis X direction, of the probe static-pressure orifices. Columns eight and nine indicate the lateral and vertical coordinates, the body-axis Y and Z directions, respectively, of the probe longitudinal centerline. These coordinates indicate the various inner flow-field surveys. The outer flow-field surveys are found on the second page of each table and at the end of each table. Columns four, five, and six of these surveys indicate the initial and final positions and incremental change, respectively, of the axial coordinate, the tunnel-axis X direction, of the probe static-pressure orifices. Columns eight and nine indicate the lateral and vertical coordinates, the tunnel-axis Y and Z directions, respectively, of the probe longitudinal centerline. Column ten indicates the azimuthal angle of each outer flow-field run, as defined in figure 4 of this volume.

3. DESCRIPTION OF DATA

The flow-field survey data for the 6-percent thick wing-body model at Mach numbers 0.925, 0.975 and 1.025 are presented in tabular form on pages 1 through 304 at the end of this volume. The heading on each page contains the test number, the part number, the Reynolds number per foot, the angle of attack of the model, the type of wing attached to the model (6-percent thick wing for this volume), and the Y and Z (or YT and ZT) coordinates at which the X (or XT) traverse is carried out. Also included are the run and survey numbers and the date on which data were recorded.

though the entire cold awork

Below the heading information are the data obtained during each test. Column one indicates the sequential indexing number for referencing data obtained during one part (POINT). Column two

indicates the location of the probe static-pressure orifices in the body-axis X direction (X) for the inner flow-field surveys or in the tunnel-axis X direction (XT) for the outer flow-field surveys. Columns three through seven indicate wind tunnel free-stream quantities. These are Mach number (M), velocity (VM, ft/sec), total pressure (PT, psfa), dynamic pressure (Q, psf) and total temperature (TT, OF). Columns eight through sixteen indicate local quantities which were either measured by the probe or calculated from probe measurements. Columns eight through eleven contain the local Mach number (ML), the ratio of local to free-stream velocity (VML/VM), the ratio of local to free-stream total pressure (PTL/PT), and the local pressure coefficient (CPL). For the inner flow-field surveys, columns twelve through sixteen contain the ratio of local velocity components in the body-axis X, Y, and Z directions, respectively, to the free-stream velocity (UL/VM, VL/VM, and WL/VM, respectively) and the local upwash and sidewash angles (AAL and SWL, respectively) referenced to body-axis coordinates. For the outer flow-field surveys, columns twelve through sixteen contain these same local quantities as determined in the tunnel-axis coordinate system. The positive sense of the velocity components is along the positive X, Y, and Z directions. A positive local upwash angle indicates downward flow away from the wing-body combination, the positive Z or ZT direction, see figure 5. A positive local sidewash angle indicates flow along the positive Y or YT axis, see figure 5.

TABLE I. - FLOW-FIELD SURVEY TESTS

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6-Percent Thick Wing Body Model

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119	956		10.333	saliset Spyle (Spyle State) in Colore Aug 19	0.666		2.000	-5.000	
120	961	J.,	10.000		0.333		1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1	-3.000	
121	962	Water Market Comment						-2.000	
122	963							-1.000	- Januari
123	957	The second second second	11,000		10.666	de la como en	4.000	-5.000	
124	964				0.333	and project the		-3.000	
125	965					and the factor of the state of	1	-2.000	
126	. 966		arte are de la como cardo de la	alian makeling a later on the	eron de la companya d	ere subsequently		-1.000	
127	958		13,000		0.666		7.000	-5.000	
128	967				0.333		1 - 1 - 1 - 1 - 1 - 1 - 1 - 1 - 1 - 1 -	-3.000	
129	968							-2.000	
130	969			e har on the said				-1.000	
131	970		11.000		1.333		-4.000	-5.000	
132	971					N. 2 () 1 1 1		-3.000	
133	972				aman, Alahasa Alamanan			-1.000	
- 134	973		13.666				-7.000	-5.000	
135	974		L L TR	A STATE OF THE STA				-3.000	
136	975	gr./	T		h parties			-1.000	
	T ASSET		1006	Person	C 1000 3		Tiaci		
	200	7							fi I
		***	Initial XT inches	XT	ΔXT inches		YT inches	ZT inches	degr
137, 138	977		-6,000	24.000	0.500		0.000	-14, 142	-90
139,140	978				L			-10,000	
141,142	979						14.142		
143	980		Mark Jata		2.000		-10.000	-10.000	

	TABLE	VIII.	- FLOW-F1	ELD SUR	VEYS AT	M =	0.975, α	= 2°	
1	2	3		5	6	7	8	9	1
Page No.	Part No.		Initial X inches	Final X inches	ΔX inches		Y inches	Z inches	
144	982		9,000	19.000	0.666		0.000	-5.000	
145	CONTRACTOR OF THE PARTY OF THE	7	1 3.000	1	Committee of the Commit	0.1	11 88		
146	986 987				0.333		1 1	-3.000 -2.000	
147	983	1	10.333		0.666		2,000	-5.000	
148	988		10.000		0.333	011	1 4 4 4 4	-3.000	
149	989		50.53		1 100			-2.000	
150	990		1 1		1 7 1		THE SA	-1.000	
151	984		11,000		0.666		4.000	-5.000	
152	991				0.333		1 1 20	-3.000	
153	992			0 1 1	3.333			-2.000	
154	. 993		177				1 1	-1.000	
155	985		13,000		0.666		7.000	-5.000	
156	994		13,000		0.333	2.11	1 1980	-3.000	
157	995				1 1 1		The state of the s	-2.000	
158	996						1.7 (4.5)	-1.000	
159	997		11.000		1.333		-4.000	-5.000	
160	998		1880			rri	108	-3.000	
161	999							-1.000	
162	1000		13.666				-7.000	-5.000	
163	1001				1 656		1 1 1 1 2 2 3	-3.000	
164	1002				1 11		T T DEC	-1.000	
	ta kan ang san						10 008	A A B	
S Towns of the second									
			Initial XT	XT	ΔXT		YT	ZT	
	2000		inches	inches	inches		inches	inches	ae
165,166	1004	Billion of Spatial	-6.000	24.000	0.500			-14.142	
167,168	Contract of the last of the la		1,106.4	1,100.3	-			-10.000	-
169,170 171	1006				2.000		14.142	-10.000	

1	2	3	4	5	6	7	975, α =	9	10
Page No.	Part No.		Initial X inches	x X ento	ΔX inches	31	Yinches	Z	
172	1035		9.000	19.000	0.666		0.000	-5,000	
173	1039		1		0.333			-3.000	
174	1040		The same		1		1 1	-2,000	
175	1036		10.333		0.666		2.000	-5.000	
176	1041		10.000		0.333		1 11 11 11 11	-3.000	
177	1042							-2.000	
178	1043					rel.	rena i Si per se senta di malana di Silana. Per Vine di Si	-1.000	
179	1037		11.000		0.666		4.000	-5.000	
180	1044				0.333			-3,000	
181	1045							-2.000	
182	1046							-1.000	
183	1038		13,000		0.666	100	7.000	-5.000	
184	1047				0.333			-3,000	
185	1048						17.44	-2.000	
186	1049		1 (5)		The state of			-1.000	
187	1050		11.000		1.333		-4.000	-5.000	
188	1051							-3.000	
189	1052				1 1 3 5 3 3		(3137)	-1.000	
190	1053		13.666				-7.000	-5.000	
191	1054						James	-3.000	4 16
192	1055							-1.000	
1 12 1								•	
art tra	k assu		Initial XT	Final XT	ΔXT		- YT	ZT	
	lana i		inches	inches	inches		inches	inches	deg
193, 194	1057		-6.000	24.000	0.500		0.000		
195, 196	1058				-		10.000		-4
197,198 199	1059		Latou				14.142	0.0	-13

1	2	3	4	5	6		8	9	1
Page No.	Part No.		Initial X inches	x	ΔX inches	1573 1	Y	Z inches	
200	1009		9,000	19.000	0.666	e viduorines	0.000	-5.000	
201	1013	<i>y</i>	- 1 S. K.		0.333			-3.000	
202	1014		on and the second			1	1 1000	-2.000	
203	1010		10.333	a de la composición dela composición de la composición de la composición de la composición dela composición dela composición dela composición de la composición de la composición dela composición de la composición dela c	0.666	Service Commission	2.000	-5.000	
204	1015		10.000		0.333		1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	-3.000	
205	1016		J,		1,1			-2.000	
206	1017				TYST		1 1	-1.000	
207	1011	or grand memory and	11.000		0.666		4.000	-5.000	
208	1018		12.000		0.333	al	7.000	-3.000	
209	1019	marrow (Securiosca) is			1.			-2.000	
210	.1020							-1.000	
211	1012		13.000		0.666		7.000	-5.000	
212	1021		1 (93)37		0.333			-3.000	
213	1022				J.			-2.000	
214	1023	e reason from the control		en e			The second secon	-1.000	
215 :	1024		11.000	grounder, military value of	1, 333		-4.000	-5.000	
216	1025		1.		1.333		1.000	-3.000	
. 217	1026						1 1 1	-1.000	
218	1027		13.666			4-40-2	-7.000	-5.000	
219	1028		14		9 5 0 4		1.300	-3.000	
220	1029				TYPE		T I Vaco	-1.000	
		etryona or	Initial	Final					
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221,222	1031		-6.000	24.000	0.500		0.000	-14.142	-
223,224		war and published to	T.	La Dors	1		CONTRACTOR OF THE PARTY OF THE	-10.000	
225	1033				2.000			-10.000	

No.	62 66 67 63 68 69 70 64	Initial X inches 9.000 10.333 10.000 11.000	X	ΔX inches 0.666 0.333 0.666 0.333		y inches 0.000 2.000	Z inches -5.000 -3.000 -2.000 -5.000	
No.	62 66 67 63 68 69 70	9.000 9.000 10.333 10.000	X inches	0.666 0.333		0.000	-5.000 -3.000 -2.000 -5.000	
227 10 228 10 229 10 230 10 231 10 232 10 233 10 234 10 235, 236 1072 237 10	666 667 663 668 669 770 664	10.333	19.000	0.333		+	-3.000 -2.000 -5.000	
228 10 229 10 230 10 231 10 232 10 233 10 234 10 235, 236 1072 237 10	67 63 68 69 70 64	10.333		0.333		+	-3.000 -2.000 -5.000	
228 10 229 10 230 10 231 10 232 10 233 10 234 10 235, 236 1072 237 10	67 63 68 69 70 64	10,000		0.666		2.000	-2.000 -5.000 -3.000	
229 10 230 10 231 10 232 10 233 10 234 10 235, 236 1072 237 10	63 68 69 70 64	10,000		Constitution and the Constitution of		2.000	-5.000 -3.000	
231 10 232 10 233 10 234 10 235, 236 1072 237 10	69 70 64 71			0.333			CONTRACTOR COMPANIES AND ADDRESS OF THE PARTY OF THE PART	
232 10 233 10 234 10 235, 236 1072 237 10	70 64 71	11.000		- * -		4	CONTRACTOR COMPANIES AND ADDRESS OF THE PARTY OF THE PART	
233 10 234 10 235, 236 1072 237 .10	64 71	11.000					-2.000	
234 10 235, 236 1072 237 , 10	71	11.000			Total Control	The state of the s	-1.000	
234 10 235, 236 1072 237 , 10	71			0.666		4.000	-5.000	
237 . 10	1079			0.333			-3.000	4
				4			-2.000	
238 10	80						-1.000	
	65	13.000		0.666		7.000	-5.000	
The same of the sa	81			0.333			-3.000	
	82	- 4		V		1 4	-2.000	in and a second
	83						-1.000	
	84	11.000		1.333		-4.000	-5.000	
	85	14			N Life He	1 4	-3.000	
	86						-1.000	
	87	13.666				-7.000	-5.000	
	88		-	4		1 4	-3.000	
247 10	189			V			-1.000	
	erica -	TX	Final XT	ΔXT		YT	ZT	deg
		inches	inches	inches		inches	inches	
248, 249 10	90	-6.000	24.000	0.500		0.000	-14.142	-90
250,251 10	91	1	1	1 4 1		10,000	-10.000	-45

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1000	- 1 00n				er les				
Page No.	Part No.		Initial X inches	Final X inches	∆X inches		Y inches	Z inches	
282	1246		9,000	19.000	0.666		0,000	-5.000	
283	1248				0.333			-3.000	
284	1249	Control of the control			V			-2.000	
285	1247		10.500		0.500		4.000	-5.000	
286	1250	September 1994 September 1994						-3,000	
287	1251							-2.000	
288	1252							-1,000	
289	1253		11.000		1.000	ten englise di secono	-4.000	CONTRACTOR OF THE PARTY OF THE	
290	1254							-3.000	
291	1255	Rational Property of the Park						-1.000	
1003			Initial XT	Final XT	ΔXT		- УТ	ZT	θ
			inches	inches	inches		inches	inches	degree
292,293	1257	1	-6.000	24.000	0.500			-14.142	
294, 295	1258		1 23					-10,000	
296, 297	1259		l tagefo	ti bada	n h la male		14.142	0.0	0.0
298	1260				2.000		-10.000	-10.000	-135.0

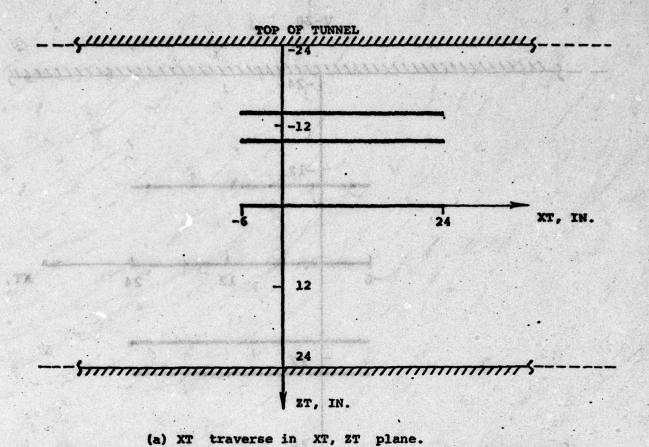
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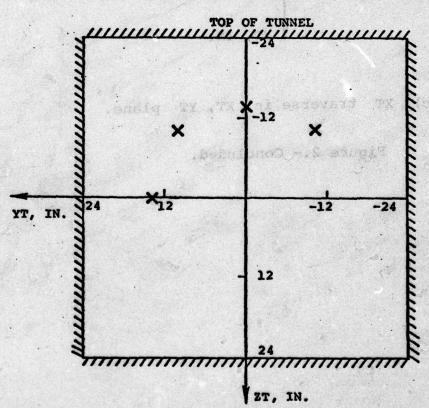
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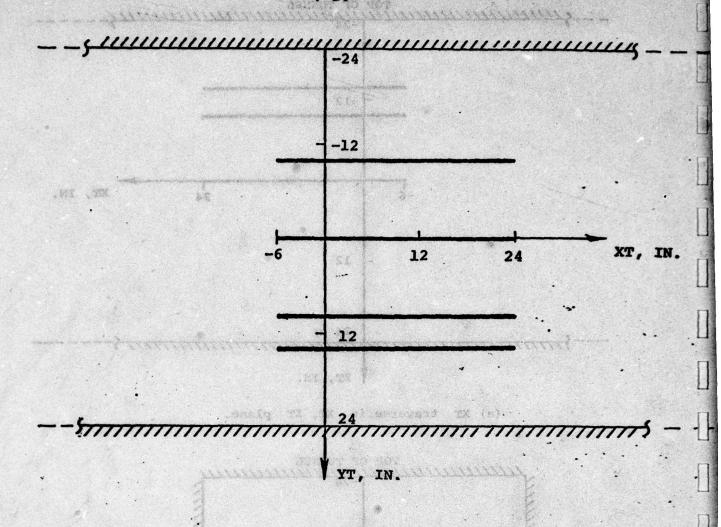
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×		X	1	XX	XX)	x x	×	X	
Page No.	Part No.		Initial X inches	Final X inches	∆X inches		Y inches	Z	
299	1266		9.000	19,000	0.666		0.000	-5.000	
300	1267		10.500	*	0.500		4.000	4	
		1					1		

		Initial XT inches	XT	∆XT inches	YT inches	ZT inches	θ degrees
301,302	1263	-6,000	24,000	0.500	0.000	-14, 142	-90.0
303,304	1264	Ψ	4	+	10.000	-10.000	-45.0





(b) XT traverse in YT, ZT plane. Figure 2.- General grids for outer flow-field surveys.



(c) XT traverse in XT, YT plane.

Figure 2.- Concluded.

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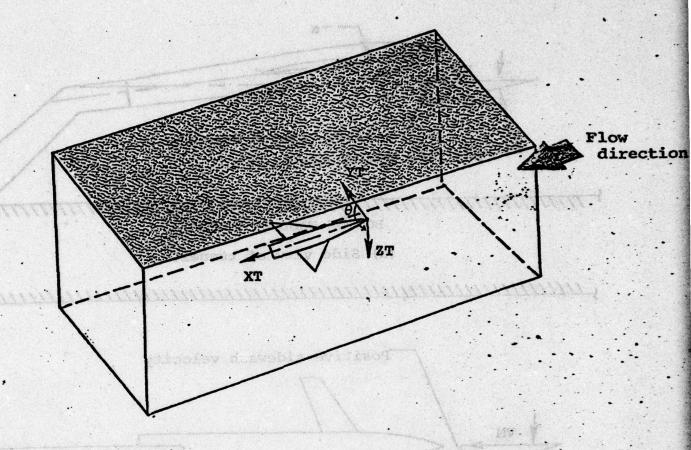
(b) 2.27 traverse to 10, 0; plane. Signer 2.4 Ceneval truck for outer these field serveys.

Figure 3 .-Wing-body combination.

Pressure orifices on body centerline Tunnel top wall direction

(a) Body-fixed system for inner flow field surveys.

Figure 4.-Coordinate systems.



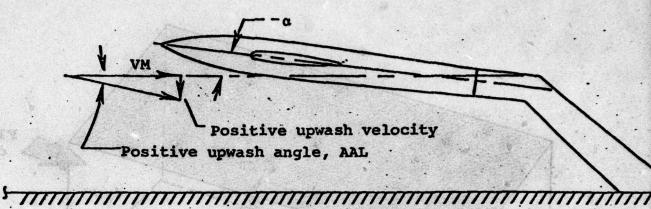
CHARACTER CONTRACTOR WATER OF THEORY CONTRACTOR OF THE CONTRACTOR

(b) Tunnel-fixed system for outer flow field surveys.

Figure 4.- Concluded.

the Plan view of tunnel from vor

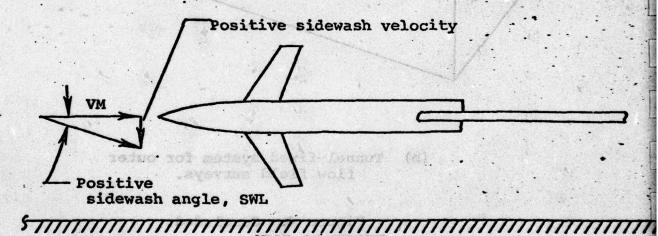
Physics N. Pictorial sign convention for application and sillowesh angles.



BOTTOM WALL OF TUNNEL

(a) Side view of tunnel.

Zummunummunummunimminummunimmi



WALL OF TUNNEL

(b) Plan view of tunnel from top.

Figure 5.-Pictorial sign convention for upwash and sidewash angles.

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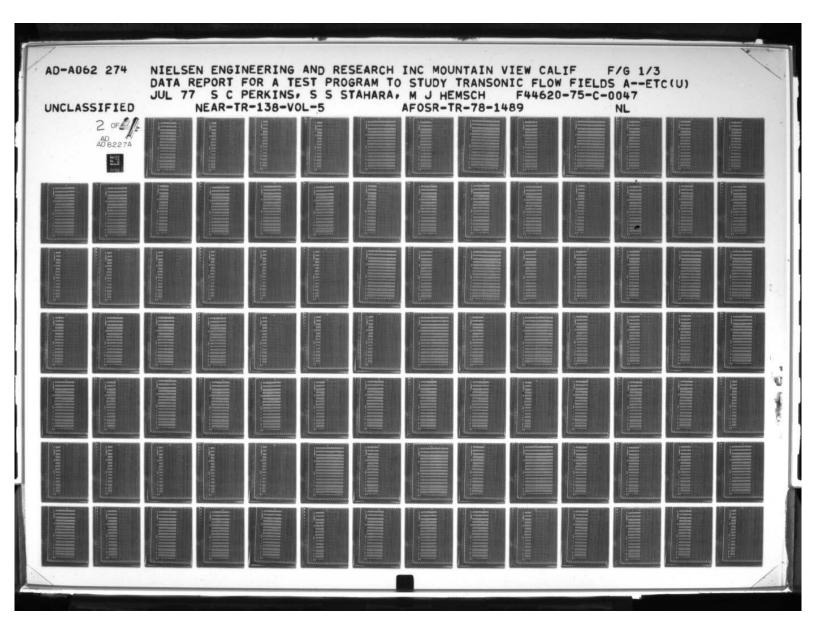
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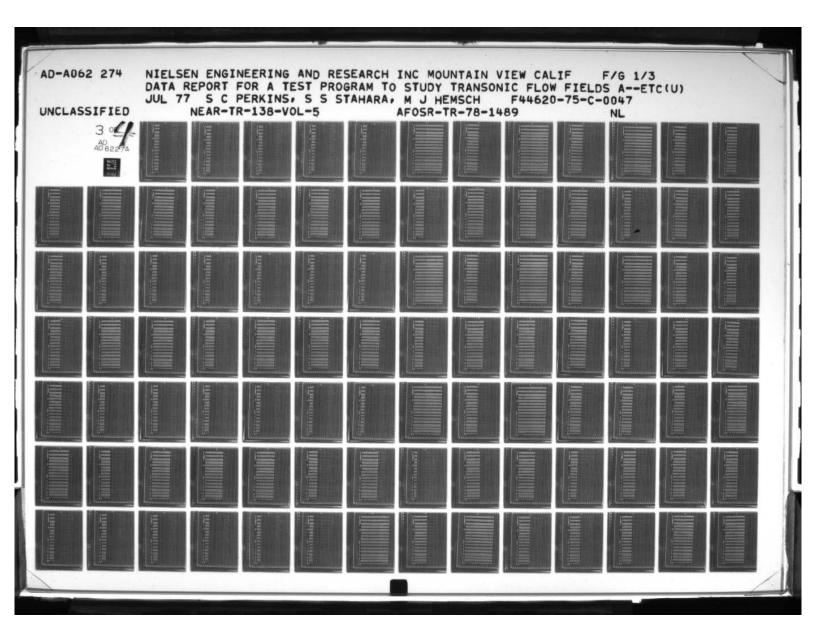
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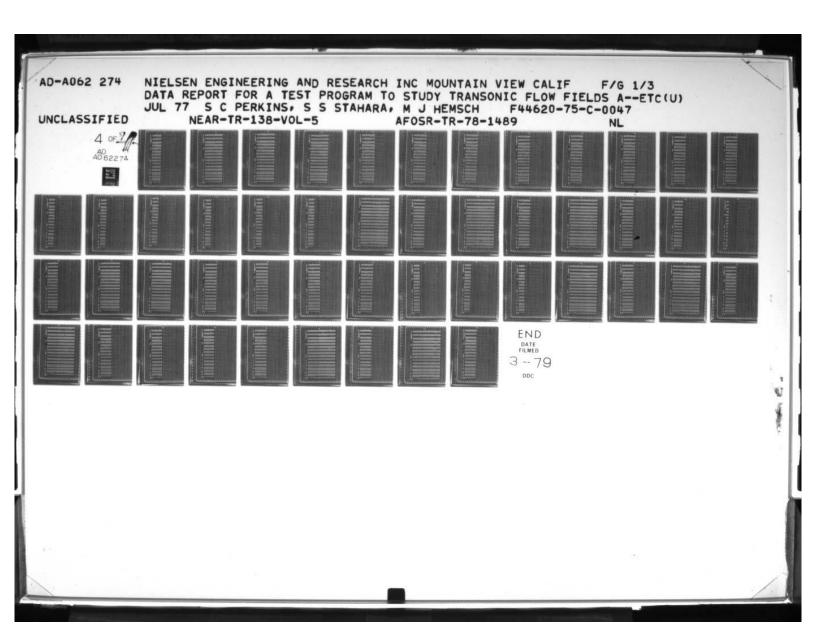
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